

Issue 07, 21 December 2016



# *European Aviation Safety Agency*

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**EASA**

**TYPE-CERTIFICATE  
DATA SHEET**

**EASA.A.089**

**PC-12**

**Pilatus Aircraft Ltd.**

P.O. Box 992  
6371 Stans  
Switzerland

For models: **PC-12**  
**PC-12/45**  
**PC-12/47**  
**PC-12/47E**

Issue 07, 21 December 2016

Issue 07, 21 December 2016

# CONTENTS

## **SECTION A: PC-12**

- AI. General
- AII. Certification Basis
- AIII. Technical Characteristics and Operational Limitations
- AIV. Operating and Service Instructions
- AV. Operational Suitability Data (OSD)

## **SECTION B: PC-12/45**

- BI. General
- BII. Certification Basis
- BIII. Technical Characteristics and Operational Limitations
- BIV. Operating and Service Instructions
- BV. Operational Suitability Data (OSD)

## **SECTION C: PC-12/47**

- CI. General
- CII. Certification Basis
- CIII. Technical Characteristics and Operational Limitations
- CIV. Operating and Service Instructions
- CV. Operational Suitability Data (OSD)

## **SECTION D: PC-12/47E**

- DI. General
- DII. Certification Basis
- DIII. Technical Characteristics and Operational Limitations
- DIV. Operating and Service Instructions
- DV. Operational Suitability Data (OSD)

## **ADMINISTRATIVE SECTION**

- I Notes for all PC-12 variants
- II Acronyms
- III Type Certificate Holder Record
- IV Change Record

Issue 07, 21 December 2016

**SECTION A: PC-12****A.I General**

Data Sheet No.: EASA.A.089

Issue: 01

Date: 23 June 2006

- |                                    |   |
|------------------------------------|---|
| 1. Type:                           | PC-12   |
| Variant:                           | PC-12   |
| 2. Airworthiness Category:         | 14 CFR Part 23 Normal Category  |
| 3. Type Certificate Holder:        | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland                                  |
| 4. Manufacturer:                   | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland                                  |
| 5. Certification Application Date: | July 10 <sup>th</sup> , 1986  |
| 6. FOCA Certification Date:        | March 30 <sup>th</sup> , 1994   |
| 7. EASA Certification Date:        | Product accepted in EU prior 28 Sept 2003   |
| 8. FOCA TCDS F-56-30:              | This TCDS replaces the Swiss TCDS F-56-30<br>Revision 10, dated December 14 <sup>th</sup> , 2005. |

**A.II Certification Basis**

- |  |  |
|--|--|
| 1. Reference Date for determining the applicable requirements: | July 10 <sup>th</sup> , 1986   |
| 2. (reserved)  |  |
| 3. (reserved)  |  |
| 4. Certification Basis:  | FOCA CRI A-1, Stage 7, February 14 <sup>th</sup> , 1997  |
| 5. Airworthiness Requirements:                                 | US 14 CFR FAR Part 23, Normal Category, including Amendments 23-1 through 23-42, effective February 4 <sup>th</sup> , 1991.  |
| 6. Requirements elected to comply:                             | FAR 23.1305(c)(3)    Amdt 23-43<br>FAR 23.1311        Amdt 23-49<br>FAR 23.1507        Amdt 23-45  |
| 7. Special Conditions:   | B-1 Stall Identification & Recovery Characteristics<br>C-1 Horizontal Tail Loads<br>C-2 Horizontal Tail Loads (Rocking Motions)<br>C-3 Dynamic Behavior of the Landing Gear<br>C-4 Seat Head Rest & Supporting Structure aft Facing Seats<br>D-1 Hinges (Strength & Rigidity)<br>D-2 Doors and Exits<br>D-3 Composite Materials for Secondary Structure<br>E-2 Composite Cowling (Toxics)<br>FOCA CQF 98-02, September 15 <sup>th</sup> , 1993 |

Issue 07, 21 December 2016

8. Exemptions: None
9. Equivalent Safety Findings: FAR 23.221(a)(2) [FOCA CQF 91-03]  
FAR 23.841(b)(6) [FOCA CQF 21-03]
10. Environmental Standards:
- Noise: US Federal Aviation Regulation Part 36, Appendix G, including Amendments 36-1 through 36-20, effective September 11, 1992.  
  
ICAO Annex 16: Environmental Protection, Second Edition, Amdt 3, effective November 17<sup>th</sup>, 1988; Volume 1, Part II, Chapter 10.
- Emissions: US Federal Aviation Regulation Part 34, (Fuel venting/emissions), effective September 10<sup>th</sup>, 1990.

## 11. Operational Suitability Certification Basis:

- MMEL: CS-GEN-MMEL, Initial Issue
- Flight Crew Data: not required (see Note 4)

12. Eligible S/N: S/N 101 to S/N 683 (except 545)

**A.III Technical Characteristics and Operational Limitations**

1. Type Design Definition: 500.00.12.001
2. Description: The PC-12 is a large single-engine turboprop multipurpose aircraft designed to perform a wide range of missions. Design features include:
- A pressurized, large volume cabin quickly convertible from all-passenger to all-cargo or a combination of passenger/cargo configurations.
  - Retractable landing gear with trailing-link main landing gear capable of grass and unprepared field operations.
  - Complete icing protection for flight into known icing conditions (see Note A.IV.5)
3. Dimensions:
- |                  |                      |                          |
|------------------|----------------------|--------------------------|
| Main Wing Span:  | 16'230 mm            | (53 ft 3 in)             |
| Length:          | 14'408 mm            | (47 ft 3 in)             |
| Height:          | 4'260 mm             | (14 ft 0 in)             |
| Total Wing Area: | 25.81 m <sup>2</sup> | (277.8 ft <sup>2</sup> ) |
4. Engines:
- Model: 1 Pratt & Whitney Canada (PWC) PT6A-67B turboprop engine, flat rated at 1200 shp for takeoff.

Issue 07, 21 December 2016

Type Certificate:

Transport Canada Type Approval No. E21

## 5. Engine Limits:

Operating Conditions	Shaft (shp.)	N1 Gas Generator Speed (%)	Torque PSI [ft-lbs] (kgm)	Prop. shaft Speed (r.p.m.)	Maximum Permissible Interstage Temperature (°C)
Takeoff	1200	104	44.34 [3708] (512.7)	1700	800
Max. continuous Max. climb Max. cruise	1000	104	36.95 [3090] (427.2)	1700	760
Normal Climb Normal Cruise	As per Aircraft Flight Manual charts				
Starting Limits (5 sec.)	-	-	-	-	1000
Transient (20 sec.)	-	104	61.00 [5100] (705.1)	1870	870

Note: 100% Gas Generator Speed = 37'468 RPM

## Oil Temperature:

- Starting: - 40°C (min.)
- Idle: - 40°C ÷ 110°C
- Transient: - 40°C ÷ 110°C
- Take-off: +10°C ÷ 110°C
- Max. Continuous: +10°C ÷ 105°C
- Max. Reverse: +10°C ÷ 105°C

## 6. Propeller:

Model: 1 Hartzell HC-E4A-3D/E10477K  
or HC-E4A-3D/E10477SK

Type Certificate: TC No. P10NE issued by the FAA.

Number of blades: 4 (Aluminum)

- 6.1. Sense of Rotation: Propeller rotates Clockwise in view of flight direction
- 6.2. Diameter: 2,670 mm
- 6.3. Pitch: Nominal pitch angle at 1,067 m (42") station
- Minimum on ground: 17°
  - Minimum in flight: 6°
  - Reverse (negative): -17.50° ± 0,5°
  - Feathered: 79.60° ± 0,5°
  - Fine Pitch: 19° ± 0,2°

Issue 07, 21 December 2016

## 6.4. Propeller Limits:

- Diameter: 104 in (2.642 m) to 105 in (2.667 m) cropping of blade tips not permitted.
- Stabilized ground operations between 350 and 950 rpm are prohibited.
- Propeller blade life limit on condition.

## 7. Fluids:

## 7.1. Fuel:

- Refer to the latest revision of Pratt & Whitney Service Bulletin No. 14004 (including JET A, JET A-1, JET B, JP4).
- Fuel Anti-Ice Additive compliant with Specification MIL-DTL-27686 or MIL-DTL-85470 must be used for all flight operations in ambient temperatures below 0°C.

## 7.2. Oil:

- Synthetic turbine oil conforming to PWA 521, Type II. For acceptable oil brands see Pratt & Whitney Service Bulletin No. 14001.

## 8. Fuel capacities

## 8.1. Fuel:

Total:	1540 lt (406.8 US Gal)
Usable:	1522 lt (402.1 US Gal)
Unusable:	19.6 kg (43.2 lbs) S/N 101 to S/N 140 (incl.) 14.9 kg (32.9 lbs) S/N 141 and up

## 8.2. Oil:

Total:	13.6 lt (3.6 US Gal)
Usable quantity:	5.68 lt (1.5 US Gal)

## 9. Air Speeds:

		<b>KCAS</b>
VMO	(maximum operating speed)	240
MMO	(maximum operating Mach number)	0.48
VD	(maximum diving speed)	280
MD	(maximum operating Mach number)	0.60
VA	(maneuvering speed)	170
VO	(max. maneuvering operating speed)	at 4100 kg 154 at 3200 kg 136 at 2600 kg 123
VFE	(max. flap extended speed)	up to 15° 165 above 15° 130
VFO	(max. flap operating speed)	up to 15° 165 above 15° 130
VLO	(maximum landing gear operating speed)	180
VLE	(maximum landing gear extended speed)	240

Issue 07, 21 December 2016

10. Maximum Operating Altitude: 9144 m / 30000 ft
11. Operational Capabilities: IFR Day/Night; VFR Day/Night
12. Maximum Weight:
- |               |         |            |
|---------------|---------|------------|
| Taxi and ramp | 4120 kg | (9083 lbs) |
| Take-off      | 4100 kg | (9039 lbs) |
| Landing       | 4100 kg | (9039 lbs) |
| Zero fuel     | 3700 kg | (8157 lbs) |
13. Centre of Gravity Range: Straight line variation between limits given
- | <b>Weight</b>           | <b>From</b>       | <b>To</b>         |
|-------------------------|-------------------|-------------------|
| 4100 kg ( 9039 lbs)     | 5.847 m (230.18") | 6.137 m (241.61") |
| 3700 kg ( 8157 lbs)     | 5.689 m (223.99") | 6.163 m (242.73") |
| 3600 kg ( 7937 lbs)     | 5.684 m (223.78") | 6.172 m (242.99") |
| 3000 kg ( 6614 lbs)     | 5.633 m (221.85") | 6.172 m (242.99") |
| 2700 kg ( 5953 lbs)     | 5.607 m (220.75") | 5.880 m (231.50") |
| 2550 kg ( 5622 lbs)     | 5.607 m (220.75") | 5.728 M (225.47") |
| Empty Weight C.G. Range | None              |                   |
14. Datum: 3.000 m (118.11") forward of Frame 10 (foremost cabin frame = firewall)
15. Mean Aerodynamic Chord (MAC): 1.710 m (67.32")
16. Leveling Means: Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6
17. Minimum Flight Crew: 1 Pilot
18. Maximum Passenger Seating Capacity: 9 PAX excluding pilot seats.  
Refer to the FOCA/EASA Approved Flight Manual, Section 6, for passengers and flight crew loading instructions and approved configurations.
19. Exit: Nb. and Type: 3 exits (forward cabin LH door, cargo LH door and one RH over wing emergency exit).
20. Baggage / Cargo Loading: Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6.
21. Wheels and Tyres:
- 21.1. Wheels: Nose Landing Gear: BF Goodrich 3-1501  
Main Landing Gear: BF Goodrich 3-1543-1

Issue 07, 21 December 2016

21.2. Tyres:

	Dimensions	Ply Rating	Speed Rating
Nose Landing Gear:	17.5x6.25-6	8 (PR)	160 (MPH)
Main Landing Gear:	8.50x10	8 (PR)	160 (MPH)





Issue 07, 21 December 2016

**SECTION B: PC-12/45****B.I General**

Data Sheet No.: EASA.A.089	Issue: 01	Date: 23 June 2006
1. Type: Variant:	PC-12 PC-12/45	
2. Airworthiness Category:	14 CFR Part 23 Normal Category	
3. Type Certificate Holder:	Pilatus Aircraft Ltd. P.O. Box 992, 6371 Stans Switzerland	
4. Manufacturer:	Pilatus Aircraft Ltd. P.O. Box 992, 6371 Stans Switzerland	
5. Certification Application Date:	June 6 <sup>th</sup> , 1995	
6. FOCA Certification Date:	June 4 <sup>th</sup> , 1996	
7. EASA Certification Date:	Product transferred to EASA, 28 Sept 2003	
8. FOCA TCDS F-56-30:	This TCDS replaces the Swiss TCDS F-56-30 Revision 10, dated December 14 <sup>th</sup> , 2005.	

**B.II Certification Basis**

1. Reference Date for determining the applicable requirements:	June 6 <sup>th</sup> , 1995	
2. (reserved)		
3. (reserved)		
4. Certification Basis:	FOCA CRI A-2, Stage 5, February 14 <sup>th</sup> , 1997	
5. Airworthiness Requirements:	US 14 CFR FAR Part 23, Normal Category, including Amendments 23-1 through 23-42, effective February 4 <sup>th</sup> , 1991.	
6. Requirements elected to comply:	FAR 23.49(c)	Amdt 23-44
	FAR 23.479(b)(c)	Amdt 23-45
	FAR 23.562(d)	Amdt 23-44
	FAR 23.1305(c)(3)	Amdt 23-43
	FAR 23.1311	Amdt 23-49
	FAR 23.1507	Amdt 23-45
7. Special Conditions:	B-1 Stall Identification & Recovery Characteristics C-1 Horizontal Tail Loads C-2 Horizontal Tail Loads (Rocking Motions) C-3 Dynamic Behavior of the Landing Gear C-4 Seat Head Rest & Supporting Structure aft Facing Seats D-1 Hinges (Strength & Rigidity) D-2 Doors and Exits D-3 Composite Materials for Secondary Structure E-2 Composite Cowling (Toxics) FOCA CQF 98-02, September 15 <sup>th</sup> , 1993	

Issue 07, 21 December 2016

8. Exemptions: None
9. Equivalent Safety Findings: FAR 23.221(a)(2) [FOCA CQF 91-04]  
FAR 23.841(b)(6) [FOCA CQF 21-03]
10. Environmental Standards:
- Noise: US Federal Aviation Regulation Part 36, Appendix G, including Amendments 36-1 through 36-20, effective September 11, 1992.  
  
ICAO Annex 16: Environmental Protection, Second Edition, Amdt 3, effective November 17<sup>th</sup>, 1988; Volume 1, Part II, Chapter 10.
- Emissions: US Federal Aviation Regulation Part 34, (Fuel venting/emissions), effective September 10<sup>th</sup>, 1990.
11. Operational Suitability Certification Basis:
- MMEL: CS-GEN-MMEL, Initial Issue
- Flight Crew Data: not required (see Note 4)
12. Eligible S/N: S/N 101 to S/N 683 (except 545)

### **B.III Technical Characteristics and Operational Limitations**

1. Type Design Definition: 500.00.12.009
2. Description: The PC-12/45 is a large single-engine turboprop multipurpose aircraft designed to perform a wide range of missions. Design features include:
- A pressurized, large volume cabin quickly convertible from all-passenger to all-cargo or a combination of passenger/cargo configurations.
  - Retractable landing gear with trailing-link main landing gear capable of grass and unprepared field operations.
  - Complete icing protection for flight into known icing conditions (see Note B.IV.5)
3. Dimensions:
- |                  |  |
|------------------|--|
| Main Wing Span:  | 16'230 mm (53 ft 3 in) S/N 101-683 (excl. 545) |
|                  | 16'280 mm (53 ft 5 in) S/N 684 and up          |
| Length:          | 14'408 mm (47 ft 3 in)                         |
| Height:          | 4'260 mm (14 ft 0 in)                          |
| Total Wing Area: | 25.81 m <sup>2</sup> (277.8 ft <sup>2</sup> )  |
4. Engines:
- Model: 1 Pratt & Whitney Canada (PWC) PT6A-67B turboprop engine, flat rated at 1200 shp for takeoff.
- Type Certificate: Transport Canada Type Approval No. E21

Issue 07, 21 December 2016

## 5. Engine Limits:

Operating Conditions	Shaft (shp.)	N1 Gas Generator Speed (%)	Torque PSI [ft-lbs] (kgm)	Prop. shaft Speed (r.p.m.)	Maximum Permissible Interstage Temperature (°C)
Takeoff	1200	104	44.34 [3708] (512.7)	1700	800
Max. continuous Max. climb Max. cruise	1000	104	36.95 [3090] (427.2)	1700	760
Normal Climb Normal Cruise	As per Aircraft Flight Manual charts				
Starting Limits (5 sec.)	-	-	-	-	1000
Transient (20 sec.)	-	104	61.00 [5100] (705.1)	1870	870

Note: 100% Gas Generator Speed = 37'468 RPM

## Oil Temperature:

- Starting: - 40°C (min.)
- Idle: - 40°C ÷ 110°C
- Transient: - 40°C ÷ 110°C
- Take-off: +10°C ÷ 110°C
- Max. Continuous: +10°C ÷ 105°C
- Max. Reverse: +10°C ÷ 105°C

## 6. Propeller:

Model: 1 Hartzell HC-E4A-3D/E10477K  
or HC-E4A-3D/E10477SK

Type Certificate: TC No. P10NE issued by the FAA.

Number of blades: 4 (Aluminum)

6.1. Sense of Rotation: Propeller rotates Clockwise in view of flight direction

6.2. Diameter: 2,670 mm

6.3. Pitch: Nominal pitch angle at 1,067 m (42") station

- Minimum on ground: 17°
- Minimum in flight: 6°
- Reverse (negative): -17.50° ± 0,5°
- Feathered: 79.60° ± 0,5°
- Fine Pitch: 19° ± 0,2°

6.4. Propeller Limits:

- Diameter: 104 in (2.642 m) to 105 in (2.667 m) cropping of blade tips not permitted.

Issue 07, 21 December 2016

- Stabilized ground operations between 350 and 950 rpm are prohibited.
- Propeller blade life limit on condition.

## 7. Fluids:

## 7.1. Fuel:

- Refer to the latest revision of Pratt & Whitney Service Bulletin No. 14004 (including JET A, JET A-1, JET B, JP4).
- Fuel Anti-Ice Additive compliant with Specification MIL-DTL-27686 or MIL-DTL-85470 must be used for all flight operations in ambient temperatures below 0°C.

## 7.2. Oil:

- Synthetic turbine oil conforming to PWA 521, Type II. For acceptable oil brands see Pratt & Whitney Service Bulletin No. 14001.

## 8. Fuel capacities

## 8.1. Fuel:

Total:	1540 lt	(406.8 US Gal)
Usable:	1522 lt	(402.1 US Gal)
Unusable:	19.6 kg	(43.2 lbs) S/N 101 to S/N 140 (incl.)
	14.9 kg	(32.9 lbs) S/N 141 and up

## 8.2. Oil:

Total:	13.6 lt	(3.6 US Gal)
Usable quantity:	5.68 lt	(1.5 US Gal)

## 9. Air Speeds:

		<b>KCAS</b>
VMO	(maximum operating speed)	240
MMO	(maximum operating Mach number)	0.48
VD	(maximum diving speed)	290
MD	(maximum operating Mach number)	S/N 101-683 (excl 545) 0.62
		S/N 684 and up 0.58
VA	(maneuvering speed)	170
VO	(max. maneuvering operating speed)	at 4500 kg 161
		at 4100 kg 154
		at 3200 kg 136
		at 2600 kg 123
VFE	(max. flap extended speed)	up to 15° 165
		above 15° 130
VFO	(max. flap operating speed)	up to 15° 165
		above 15° 130
VLO	(maximum landing gear operating speed)	180
VLE	(maximum landing gear extended speed)	240

Issue 07, 21 December 2016

10. Maximum Operating Altitude:	9144 m / 30000 ft		
11. Operational Capabilities:	IFR Day/Night; VFR Day/Night		
12. Maximum Weight:			
Taxi and ramp	4520 kg	(9965 lbs)	
Take-off	4500 kg	(9921 lbs)	
Landing	4500 kg	(9921 lbs)	
Zero fuel	4100 kg	(9039 lbs)	
13. Centre of Gravity Range:	Straight line variation between limits given (!)		
<b>Weight</b>	<b>From</b>	<b>To</b>	
4500 kg ( 9921 lbs)	5.898 m (232.20")	6.120 m (240.94")	
3700 kg ( 8157 lbs)	5.693 m (224.13")		
3600 kg ( 7937 lbs)		6.172 m (242.99")	
3000 kg ( 6614 lbs)		6.172 m (242.99")	
2600 kg ( 5732 lbs)	5.607 m (220.75")	5.728 M (225.47")	
Empty Weight C.G. Range	None		
14. Datum:	3.000 m (118.11") forward of Frame 10 (foremost cabin frame = firewall)		
15. Mean Aerodynamic Chord (MAC):	1.710 m (67.32")		
16. Leveling Means:	Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6		
17. Minimum Flight Crew:	1 Pilot		
18. Maximum Passenger Seating Capacity:	9 PAX excluding pilot seats. Refer to the FOCA/EASA Approved Flight Manual, Section 6, for passengers and flight crew loading instructions and approved configurations.		
19. Exit: Nb. and Type:	3 exits (forward cabin LH door, cargo LH door and one RH over wing emergency exit).		
20. Baggage / Cargo Loading:	Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6.		
21. Wheels and Tyres:			
21.1. Wheels:	Nose Landing Gear: BF Goodrich 3-1501 Main Landing Gear: BF Goodrich 3-1543-1		
21.2. Tyres:	Dimensions	Ply Rating	Speed Rating
Nose Landing Gear:	17.5x6.25-6	8 (PR)	160 (MPH)
Main Landing Gear:	8.50x10	10 (PR)	160 (MPH)

Issue 07, 21 December 2016

#### **B.IV Operating and Service Instructions**

1. Aircraft Flight Manual (AFM):

Airplane operation must be in accordance with the FOCA/EASA approved "Pilot Operating Handbook" (POH) and AFM supplements as define below:

- |   |  |
|---|--|
| a. S/N 101-400<br>Except 321            | Pilatus Report PC-12 no. 01973-001<br>March 30 <sup>th</sup> , 1994 and later approved revisions.<br>(PC-12/45 data contained in AFMS No.8.) |
| b. S/N 321, 401 up to 683<br>Except 545 | Pilatus Report PC-12/45 no. 02211<br>July 14 <sup>th</sup> , 2001 and later approved revisions.  |

2. Aircraft Maintenance Manual (AMM):

Airplane maintenance must be in accordance with the document as define below:

All PC-12 S/N up to 999 Pilatus Report no. 02049.

3. Structural Repair Manual (SRM):

Airplane Repairs must be in accordance with the document as define below:

All PC-12 S/N up to 999 Pilatus Report no. 02050.

4. Service Bulletins (SBs):

All Pilatus PC-12 Bulletin are listed in the following document:

All PC-12 S/N Pilatus Report no. 02086.

5. Flight into icing conditions:

PC-12/45 variant may be operated in known icing conditions. For aircraft MSN 101 through MSN 128 Pilatus Service Bulletin No 30-001 must be executed.

#### **B.V Operational Suitability Data (OSD)**

- |   |   |
|---|---|
| 1. Master Minimum Equipment List (MMEL) | Pilatus Report PC-12 No 02395, latest approved revision |
| 2. Flight Crew Data (FCD)               | not required (see Note 4)                               |

Issue 07, 21 December 2016

**SECTION C: PC-12/47****C.I General**

Data Sheet No.: EASA.A.089

Issue: 02

Date: 04 July 2006

- |                                    |   |
|------------------------------------|---|
| 1. Type:<br>Variant:               | PC-12<br>PC-12/47   |
| 2. Airworthiness Category:         | 14 CFR Part 23 Normal Category  |
| 3. Type Certificate Holder:        | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland                                  |
| 4. Manufacturer:                   | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland                                  |
| 5. Certification Application Date: | September 22 <sup>nd</sup> , 2004   |
| 6. FOCA Certification Date:        | December 14 <sup>th</sup> , 2005  |
| 7. EASA Certification Date:        | Product transferred to EASA, 23 June 2006   |
| 8. FOCA TCDS F-56-30:              | This TCDS replaces the Swiss TCDS F-56-30<br>Revision 10, dated December 14 <sup>th</sup> , 2005. |

**C.II Certification Basis**

- |  |  |              |            |                  |            |               |            |                   |            |             |            |             |            |
|--|--|--------------|------------|------------------|------------|---------------|------------|-------------------|------------|-------------|------------|-------------|------------|
| 1. Reference Date for determining the applicable requirements: | September 22 <sup>nd</sup> , 2004  |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 2. (reserved)  |  |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 3. (reserved)  |  |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 4. Certification Basis:  | FOCA CRI A-1, Stage 2, November 3 <sup>rd</sup> , 2005   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 5. Airworthiness Requirements:                                 | US 14 CFR FAR Part 23, Normal Category, including Amendments 23-1 through 23-42, effective February 4 <sup>th</sup> , 1991.  |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 6. Requirements elected to comply:                             | <table border="0"> <tr> <td>FAR 23.49(c)</td> <td>Amdt 23-44</td> </tr> <tr> <td>FAR 23.479(b)(c)</td> <td>Amdt 23-45</td> </tr> <tr> <td>FAR 23.562(d)</td> <td>Amdt 23-44</td> </tr> <tr> <td>FAR 23.1305(c)(3)</td> <td>Amdt 23-43</td> </tr> <tr> <td>FAR 23.1311</td> <td>Amdt 23-49</td> </tr> <tr> <td>FAR 23.1507</td> <td>Amdt 23-45</td> </tr> </table>  | FAR 23.49(c) | Amdt 23-44 | FAR 23.479(b)(c) | Amdt 23-45 | FAR 23.562(d) | Amdt 23-44 | FAR 23.1305(c)(3) | Amdt 23-43 | FAR 23.1311 | Amdt 23-49 | FAR 23.1507 | Amdt 23-45 |
| FAR 23.49(c)   | Amdt 23-44   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| FAR 23.479(b)(c)   | Amdt 23-45   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| FAR 23.562(d)  | Amdt 23-44   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| FAR 23.1305(c)(3)  | Amdt 23-43   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| FAR 23.1311  | Amdt 23-49   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| FAR 23.1507  | Amdt 23-45   |              |            |                  |            |               |            |                   |            |             |            |             |            |
| 7. Special Conditions:   | B-1 Stall Identification & Recovery Characteristics<br>C-1 Horizontal Tail Loads<br>C-2 Horizontal Tail Loads (Rocking Motions)<br>C-3 Dynamic Behavior of the Landing Gear<br>C-4 Seat Head Rest & Supporting Structure aft Facing Seats<br>D-1 Hinges (Strength & Rigidity)<br>D-2 Doors and Exits<br>D-3 Composite Materials for Secondary Structure<br>E-2 Composite Cowling (Toxics)<br>FOCA CQF 98-02, September 15 <sup>th</sup> , 1993 |              |            |                  |            |               |            |                   |            |             |            |             |            |



Issue 07, 21 December 2016

8. Exemptions: None
9. Equivalent Safety Findings: FAR 23.221(a)(2) [FOCA CQF 91-04]  
FAR 23.841(b)(6) [FOCA CQF 21-03]
10. Environmental Standards:
- Noise: US Federal Aviation Regulation Part 36, Appendix G, including Amendments 36-1 through 36-27, effective June 09<sup>th</sup>, 2005.
- ICAO Annex 16: Environmental Protection, Third Edition, Amdt 7, effective March 21<sup>st</sup>, 2002; Volume 1, Part II, Chapter 6 and 10.
- Emissions: US Federal Aviation Regulation Part 34, (Fuel venting/emissions), effective September 10<sup>th</sup>, 1990.
11. Operational Suitability Certification Basis:
- MMEL: CS-GEN-MMEL, Initial Issue
- Flight Crew Data: not required (see Note 4)
12. Eligible S/N: S/N 684 up to 999

### **C.III Technical Characteristics and Operational Limitations**

1. Type Design Definition: 500.00.12.015
2. Description: The PC-12/47 is a large single-engine turboprop multipurpose aircraft designed to perform a wide range of missions. Design features include:
- A pressurized, large volume cabin quickly convertible from all-passenger to all-cargo or a combination of passenger/cargo configurations.
  - Retractable landing gear with trailing-link main landing gear capable of grass and unprepared field operations.
  - Complete icing protection for flight into known icing conditions.
3. Dimensions:
- |                  |                      |                          |
|------------------|----------------------|--------------------------|
| Main Wing Span:  | 16'280 mm            | (53 ft 5 in)             |
| Length:          | 14'408 mm            | (47 ft 3 in)             |
| Height:          | 4'260 mm             | (14 ft 0 in)             |
| Total Wing Area: | 25.81 m <sup>2</sup> | (277.8 ft <sup>2</sup> ) |
4. Engines:
- Model: 1 Pratt & Whitney Canada (PWC) PT6A-67B turboprop engine, flat rated at 1200 shp for takeoff.
- Type Certificate: Transport Canada Type Approval No. E21

Issue 07, 21 December 2016

## 5. Engine Limits:

Operating Conditions	Shaft (shp.)	N1 Gas Generator Speed (%)	Torque PSI [ft-lbs] (kgm)	Prop. shaft Speed (r.p.m.)	Maximum Permissible Interstage Temperature (°C)
Takeoff	1200	104	44.34 [3708] (512.7)	1700	800
Max. continuous Max. climb Max. cruise	1000	104	36.95 [3090] (427.2)	1700	760
Normal Climb Normal Cruise	As per Aircraft Flight Manual charts				
Starting Limits (5 sec.)	-	-	-	-	1000
Transient (20 sec.)	-	104	61.00 [5100] (705.1)	1870	870

Note: 100% Gas Generator Speed = 37'468 RPM

## Oil Temperature:

- Starting: - 40°C (min.)
- Idle: - 40°C ÷ 110°C
- Transient: - 40°C ÷ 110°C
- Take-off: +10°C ÷ 110°C
- Max. Continuous: +10°C ÷ 105°C
- Max. Reverse: +10°C ÷ 105°C

## 6. Propeller:

Model: 1 Hartzell HC-E4A-3D/E10477K  
or HC-E4A-3D/E10477SK

Type Certificate: TC No. P10NE issued by the FAA.

Number of blades: 4 (Aluminum)

6.1. Sense of Rotation: Propeller rotates Clockwise in view of flight direction

6.2. Diameter: 2,670 mm

6.3. Pitch: Nominal pitch angle at 1,067 m (42") station

- Minimum on ground: 17°
- Minimum in flight: 6°
- Reverse (negative): -17.50° ± 0,5°
- Feathered: 79.60° ± 0,5°
- Fine Pitch: 19° ± 0,2°

6.4. Propeller Limits:

- Diameter: 104 in (2.642 m) to 105 in (2.667 m) cropping of blade tips not permitted.

Issue 07, 21 December 2016

- Stabilized ground operations between 350 and 950 rpm are prohibited.
- Propeller blade life limit on condition.

## 7. Fluids:

## 7.1. Fuel:

- Refer to the latest revision of Pratt & Whitney Service Bulletin No. 14004 (including JET A, JET A-1, JET B, JP4).
- Fuel Anti-Ice Additive compliant with Specification MIL-DTL-27686 or MIL-DTL-85470 must be used for all flight operations in ambient temperatures below 0°C.

## 7.2. Oil:

- Synthetic turbine oil conforming to PWA 521, Type II. For acceptable oil brands see Pratt & Whitney Service Bulletin No. 14001.

## 8. Fuel capacities

## 8.1. Fuel:

Total:	1540 lt	(406.8 US Gal)
Usable:	1522 lt	(402.1 US Gal)
Unusable:	14.9 kg	(32.9 lbs)

## 8.2. Oil:

Total:	13.6 lt	(3.6 US Gal)
Usable quantity:	5.68 lt	(1.5 US Gal)

## 9. Air Speeds:

		<b>KCAS</b>
VMO	(maximum operating speed)	240
MMO	(maximum operating Mach number)	0.48
VD	(maximum diving speed)	290
MD	(maximum operating Mach number)	0.58
VA	(maneuvering speed)	170
VO	(max. maneuvering operating speed)	at 4740 kg 166
		at 4500 kg 161
		at 4100 kg 154
		at 3200 kg 136
		at 2600 kg 123
VFE	(max. flap extended speed)	up to 15° 165
		above 15° 130
VFO	(max. flap operating speed)	up to 15° 165
		above 15° 130
VLO	(maximum landing gear operating speed)	180
VLE	(maximum landing gear extended speed)	240

Issue 07, 21 December 2016

10. Maximum Operating Altitude:	9144 m / 30000 ft		
11. Operational Capabilities:	IFR Day/Night; VFR Day/Night		
12. Maximum Weight:			
Taxi and ramp	4760 kg	(10494 lbs)	
Take-off	4740 kg	(10450 lbs)	
Landing	4500 kg	( 9921 lbs)	
Zero fuel	4100 kg	( 9039 lbs)	
13. Centre of Gravity Range:	Straight line variation between limits given (!)		
	<b>Weight</b>	<b>From</b>	<b>To</b>
	4740 kg (10450 lbs)	5.898 m (232.20")	6.107 m (240.43")
	4500 kg ( 9921 lbs)	5.898 m (232.20")	6.120 m (240.94")
	3700 kg ( 8157 lbs)	5.693 m (224.13")	∣
	3600 kg ( 7937 lbs)	∣	6.172 m (242.99")
	3000 kg ( 6614 lbs)	∣	6.172 m (242.99")
	2600 kg ( 5732 lbs)	5.607 m (220.75")	5.728 M (225.47")
	Empty Weight C.G. Range	None	
14. Datum:	3.000 m (118.11") forward of Frame 10 (foremost cabin frame = firewall)		
15. Mean Aerodynamic Chord (MAC):	1.710 m (67.32")		
16. Leveling Means:	Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6		
17. Minimum Flight Crew:	1 Pilot		
18. Maximum Passenger Seating Capacity:	9 PAX excluding pilot seats. Refer to the FOCA/EASA Approved Flight Manual, Section 6, for passengers and flight crew loading instructions and approved configurations.		
19. Exit: Nb. and Type:	3 exits (forward cabin LH door, cargo LH door and one RH over wing emergency exit).		
20. Baggage / Cargo Loading:	Refer to the "Pilot's Operating Handbook and FOCA/EASA Approved Flight Manual", Section 6.		
21. Wheels and Tyres:			
21.1. Wheels:	Nose Landing Gear: BF Goodrich 3-1501 Main Landing Gear: BF Goodrich 3-1543-1		
21.2. Tyres:	Dimensions	Ply Rating	Speed Rating
Nose Landing Gear:	17.5x6.25-6	8 (PR)	160 (MPH)
Main Landing Gear:	8.50x10	10 (PR)	160 (MPH)



Issue 07, 21 December 2016

**SECTION D: PC-12/47E****D.I General**

Data Sheet No.: EASA.A.089

Issue: 06

Date: 20 October 2015 |

- |                                    |  |
|------------------------------------|--|
| 1. Type:<br>Variant:               | PC-12<br>PC-12/47E   |
| 2. Airworthiness Category:         | 14 CFR Part 23 Normal Category                                   |
| 3. Type Certificate Holder:        | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland |
| 4. Manufacturer:                   | Pilatus Aircraft Ltd.<br>P.O. Box 992, 6371 Stans<br>Switzerland |
| 5. Certification Application Date: | December 6 <sup>th</sup> , 2004                                  |
| 6. EASA Certification Date:        | 28 March 2008  |

**D.II Certification Basis**

- |   |  |
|---|--|
| 1. Reference Date for determining the applicable requirements:        | July 31 <sup>th</sup> , 2005   |
| 2. (reserved)   |  |
| 3. (reserved)   |  |
| 4. Certification Basis:   | FOCA CRI A-1, Stage 7, October 20 <sup>th</sup> , 2015   |
| 5. Airworthiness Requirements:  | US 14 CFR FAR Part 23, Normal Category, including Amendments 23-1 through 23-42, effective February 4 <sup>th</sup> , 1991.  |
| 6. Requirements elected to comply:<br>[FAR 23 Paragraph (Amdt level)] | 23.49c..... (23-44)<br>23.143 c..... (23-50)<br>23.301 ..... (23-48)<br>23.305 a ..... (23-45)<br>23.335 a,b,c,d..... (23-48)<br>23.361 a,b2 ..... (23-45)<br>23.371 a ..... (23-48)<br>23.479 b,c ..... (23-45)<br>23.561 b2-3,c3 ..... (23-48)<br>23.562 d ..... (23-44)<br>23.562 d1 ..... (23-50)<br>23.571 a ..... (23-45)<br>23.572 a1,b1 ..... (23-45)<br>23.607 c ..... (23-48)<br>23.613 ..... (23-45)<br>23.629 a,b,c,d,e,f2 .... (23-48)<br>23.773 a1-2 ..... (23-45)<br>23.1303 a,b,c,d,e,f .... (23-49) see Note 7<br>23.1303 c..... (23-62) see Note 7 |

Issue 07, 21 December 2016

23.1305 c3 .....	(23-43)
23.1307 .....	(23-49)
23.1311 .....	(23-49)
23.1322 e .....	(23-43)
23.1323 c.....	(23-49)
23.1326 a,b .....	(23-49)
23.1329 .....	(23-49)
23.1331 a,b1-2, c .....	(23-43)
23.1351 b2-3,c,c1-5,g .	(23-49)
23.1353 h .....	(23-49)
23.1357 a,e .....	(23-43)
23.1359 .....	(23-49)
23.1361 a,b,c.....	(23-49)
23.1365 b,c,c1,d,e,f.....	(23-49)
23.1431 a,b,c,d,e.....	(23-49)
23.1507 .....	(23-45)
23.1525 .....	(23-45)
23.1543 c.....	(23-50)
23.1555 e2 .....	(23-50)

7. Special Conditions:
- B-1 Stall Identification & Recovery Characteristics
  - C-1 Horizontal Tail Loads
  - C-2 Horizontal Tail Loads (Rocking Motions)
  - C-3 Dynamic Behavior of the Landing Gear
  - C-4 Seat Head Rest & Supporting Structure aft Facing Seats
  - D-1 Hinges (Strength & Rigidity)
  - D-2 Doors and Exits
  - D-3 Composite Materials for Secondary Structure
  - E-2 Composite Cowling (Toxics)
  - FOCA CQF 98-02, HIRF September 15<sup>th</sup>, 1993
  - F-1 Protection from the Effects of HIRF.
  - F-2 Protection from the Indirect Effects of Lightning.
  - F-3 Human Factors Aspects of Flight Deck Design.
  - F-9 Integrated Modular Avionics (IMA).
8. Exemptions: None
9. Equivalent Safety Findings:
- [FOCA CQF 91-04] FAR 23.221(a)(2) Spin resistance
  - [FOCA CQF 21-03] FAR 23.841(b)(6) Pressure cabin warning altitude.
  - F-10 Individual Circuit Protection with IMA System.
  - F-11 ASI Flaps Markings.
  - F-12 Probes OFF Caution.
10. Environmental Standards:
- Noise: US Federal Aviation Regulation Part 36, Appendix G, including Amendments 36-1 through 36-28, effective January 04<sup>th</sup>, 2006.  
ICAO Annex 16: Environmental Protection, Third Edition, Amdt 7, effective March 21<sup>st</sup>, 2002; Volume 1, Part II, Chapter 6 and 10.
  - Emissions: US Federal Aviation Regulation Part 34, (Fuel venting/emissions), effective September 10<sup>th</sup>, 1990.

Issue 07, 21 December 2016

#### 11. Operational Suitability Certification

Basis:

MMEL: CS-GEN-MMEL, Initial Issue

Flight Crew Data: not required (see Note 4)

12. Eligible S/N: S/N 545 and S/N 1001 and up

### **D.III Technical Characteristics and Operational Limitations**

1. Type Design Definition: 500.00.12.020

2. Description: The PC-12/47E is a large single-engine turboprop multipurpose aircraft designed to perform a wide range of missions. Design features include:

- A pressurized, large volume cabin quickly convertible from all-passenger to all-cargo or a combination of passenger/cargo configurations.
- Retractable landing gear with trailing-link main landing gear capable of grass and unprepared field operations.
- Complete icing protection for flight into known icing conditions.

3. Dimensions:

Main Wing Span:	16'280 mm	(53 ft 5 in)
Length:	14'408 mm	(47 ft 3 in)
Height:	4'260 mm	(14 ft 0 in)
Total Wing Area:	25.81 m <sup>2</sup>	(277.8 ft <sup>2</sup> )

4. Engines:

Model: 1 Pratt & Whitney Canada (PWC) PT6A-67P turboprop engine, flat rated at 1200 shp for takeoff.

Type Certificate: Transport Canada Type Approval No. E-21



Issue 07, 21 December 2016

## 5. Engine Limits:

Operating Conditions	Shaft (shp.)	N1 Gas Generator Speed (%)	Torque PSI [ft-lbs] (kgm)	Prop. shaft Speed (r.p.m.)	Maximum Permissible Interstage Temperature (°C)
Takeoff	1200	104	44.34 [3708] (512.7)	1700	850
Max. continuous Max. climb	1200	104	44.34 [3708] (512.7)	1700	820
Max. cruise	1000	104	36.95 [3090] (427.2)	1700	820
Normal Climb Normal Cruise	As per Aircraft Flight Manual charts				
Starting Limits (5 sec.)	-	-	-	-	1000
Transient (20 sec.)	-	104	61.00 [5100] (705.1)	1870	870

Note: 100% Gas Generator Speed = 37'468 RPM

## Oil Temperature:

- Starting: - 40°C (min.)
- Idle: - 40°C ÷ 110°C
- Transient: - 40°C ÷ 110°C
- Take-off: +10°C ÷ 110°C
- Max. Continuous: +10°C ÷ 105°C
- Max. Reverse: +10°C ÷ 105°C

## 6. Propeller:

Model: 1 Hartzell HC-E4A-3D/E10477SK  
 Type Certificate: TC No. P10NE issued by the FAA.  
 Number of blades: 4 (Aluminum)

or

Model: Hartzell HC-E5A-3A/ NC10245B  
 Type Certificate: TC No. P20NE issued by the FAA  
 Number of blades: 5 (Carbon Composite)

- 6.1. Sense of Rotation: Propeller rotates Clockwise in view of flight direction  
 6.2. Diameter: 2,670 mm

Issue 07, 21 December 2016

## 6.3. Pitch:

Nominal pitch angle at 1,067 m (42") station

	4-Blade propeller	5-Blade propeller
- Minimum on ground:	17°	17°
- Minimum in flight:	6°	6°
- Reverse (negative):	-17.50° ± 0,5°	-17.50° ± 0,5°
- Feathered:	79.60° ± 0,5°	80.0° ± 0,5°
- Fine Pitch	19° ± 0,2°	14.7° ± 0,2°

## 6.4. Propeller Limits:

- Diameter: 104 in (2.642 m) to 105 in (2.667 m) cropping of blade tips not permitted.
- Stabilized ground operations between 350 and 950 rpm are prohibited.
- Propeller blade life limit on condition.

## 7. Fluids:

## 7.1. Fuel:

- Refer to the latest revision of Pratt & Whitney Service Bulletin No. 14004 (including JET A, JET A-1, JET B, JP4).
- Fuel Anti-Ice Additive compliant with Specification MIL-DTL-27686 or MIL-DTL-85470 must be used for all flight operations in ambient temperatures below 0°C.

## 7.2. Oil:

- Synthetic turbine oil conforming to PWA 521, Type II. For acceptable oil brands see Pratt & Whitney Service Bulletin No. 14001.

## 8. Fuel capacities

## 8.1. Fuel:

Total:	1540 lt	(406.8 US Gal)
Usable:	1522 lt	(402.1 US Gal)
Unusable:	14.9 kg	(32.9 lbs)

## 8.2. Oil:

Total:	13.6 lt	(3.6 US Gal)
Usable quantity:	5.68 lt	(1.5 US Gal)

Issue 07, 21 December 2016

9. Air Speeds:			<b>KCAS</b>
VMO (maximum operating speed)			240
MMO (maximum operating Mach number)			0.48
VD (maximum diving speed)			290
MD (maximum operating Mach number)			0.58
VA (maneuvering speed)			170
VO (max. maneuvering operating speed)	at 4740 kg		166
	at 4500 kg		161
	at 4100 kg		154
	at 3200 kg		136
	at 2600 kg		123
VFE (max. flap extended speed)	up to 15°		165
	above 15°		130
VFO (max. flap operating speed)	up to 15°		165
	above 15°		130
VLO (maximum landing gear operating speed)			180
VLE (maximum landing gear extended speed)			240
10. Maximum Operating Altitude:	9144 m / 30000 ft		
11. Operational Capabilities:	IFR Day/Night; VFR Day/Night		
12. Maximum Weight:			
Taxi and ramp	4760 kg	(10494 lbs)	
Take-off	4740 kg	(10450 lbs)	
Landing	4500 kg	( 9921 lbs)	
Zero fuel	4100 kg	( 9039 lbs)	
13. Centre of Gravity Range:	Straight line variation between limits given (!)		
<b>Weight</b>	<b>From</b>	<b>To</b>	
4740 kg (10450 lbs)	5.898 m (232.20")	6.107 m (240.43")	
4500 kg ( 9921 lbs)	5.898 m (232.20")	6.120 m (240.94")	
3700 kg ( 8157 lbs)	5.693 m (224.13")	∣	
3600 kg ( 7937 lbs)	∣	6.172 m (242.99")	
3000 kg ( 6614 lbs)	∣	6.172 m (242.99")	
2600 kg ( 5732 lbs)	5.607 m (220.75")	5.728 M (225.47")	
Empty Weight C.G. Range	None		
14. Datum:	3.000 m (118.11") forward of Frame 10 (foremost cabin frame = firewall)		
15. Mean Aerodynamic Chord (MAC):	1.710 m (67.32")		
16. Leveling Means:	Refer to the "Pilot's Operating Handbook and EASA Approved Flight Manual", Section 6		

Issue 07, 21 December 2016

17. Minimum Flight Crew: 1 Pilot
18. Maximum Passenger Seating Capacity: 9 PAX excluding pilot seats.  
Refer to the EASA Approved Flight Manual, Section 6, for passengers and flight crew loading instructions and approved configurations.
19. Exit: Nb. and Type: 3 exits (forward cabin LH door, cargo LH door and one RH over wing emergency exit).
20. Baggage / Cargo Loading: Refer to the "Pilot's Operating Handbook and EASA Approved Flight Manual", Section 6.
21. Wheels and Tyres:
- 21.1. Wheels:
- Nose Landing Gear: BF Goodrich 3-1501  
Parker 40-423
- Main Landing Gear: BF Goodrich 3-1543-1  
Parker 40-424
- 21.2. Tyres:
- |                    | Dimensions  | Ply Rating | Speed Rating |
|--------------------|-------------|------------|--------------|
| Nose Landing Gear: | 17.5x6.25-6 | 8 (PR)     | 160 (MPH)    |
| Main Landing Gear: | 8.50x10     | 10 (PR)    | 160 (MPH)    |

#### **D.IV Operating and Service Instructions**

- Aircraft Flight Manual (AFM):  
Airplane operation must be in accordance with the EASA approved "Pilot Operating Handbook" (POH) and AFM supplements as define below:
 

S/N 545 and S/N 1001 and up	Pilatus Report PC-12/47E no. 02277, revision 6 or higher
-----------------------------	---
- Aircraft Maintenance Manual (AMM):  
Airplane maintenance must be in accordance with the document as define below:
 

S/N 545 and S/N 1001 and up	Pilatus Report no. 02300.
-----------------------------	---------------------------
- Structural Repair Manual (SRM):  
Airplane Repairs must be in accordance with the document as define below:
 

S/N 545 and S/N 1001 and up	Pilatus Report no. 02305.
-----------------------------	---------------------------

Issue 07, 21 December 2016

4. Service Bulletins (SBs):

All Pilatus PC-12 Bulletin are listed in the following document:

All PC-12 S/N

Pilatus Report no. 02086.

5. RVSM capability for PC-12/47E MSN 545 and MSN 1001 and subsequent:

All airplanes equipped with Honeywell APEX system are RVSM capable provided the operator incorporates and follows airplane flight manual supplement (AFMS) No. 4 Revision 1 dated May 28, 2009 or later version and Airplane Maintenance Manual document 02300 Revision 2 (12-B-AM-00-00-00-1), dated June 3, 2009 or later version.

6. PC-12/47E MSN 1300, MSN 1451 and subsequent: These airplanes are fitted with the Electromechanical Landing Gear (eLDG) and must be operated and maintained in accordance with the Airplane Flight Manual, document no. 02277, Supplement 11, issue dated March 2, 2012 or later revisions and Airplane Maintenance Manual, document number 02300, Airworthiness Limitations 12-B-04-00-00-00A-000B-A, dated June 4, 2012 or later EASA approved revisions.

7. The PC-12/47E aircraft is Transmitting-PED tolerant.

8. PC-12/47E MSN 1576 and subsequent as well as aircraft which have SB 34-042 (Introduction of the L3 ESIS with linked independent magnetometer) embodied, can have the standby magnetic compass removed. In this case the aircraft is compliant with FAR 23.1303(c) at 23-62.

9. PC-12/47E MSN 1576 and subsequent: These airplanes are eligible to be fitted with the Hartzell 5-Blade Composite Propeller. The aircraft must be operated in accordance with the Airplane Flight Manual, document no. 02277 revision 15 or later versions and Airplane Maintenance Manual, document number 02300 Revision 14, dated November 06, 2015 or later EASA approved revisions.

10. Parker Wheels have a STC SA1376CH, but have been approved for TC inclusion via a reclassified Pilatus Minor change in Jul 2010. The Parker Wheels is now the only installation used in production aircraft since MSN 1231.

#### **D.V Operational Suitability Data (OSD)**

- |   |   |
|---|---|
| 1. Master Minimum Equipment List (MMEL) | Pilatus Report PC-12 No 02395, latest approved revision |
| 2. Flight Crew Data (FCD)               | not required (see Note 4)                               |

Issue 07, 21 December 2016

## **ADMINISTRATIVE SECTION**

### **I. Notes for all PC-12 variants**

#### 1. Requirements for the issue of the C. of A.

- The minimum required equipment as prescribed in the applicable airworthiness regulations must be installed on the individual aircraft for certification.
- Current weight and balance data, a list of equipment included in the certification empty weight and loading information when necessary must be provided for each aircraft when the C.o.A. will be issued.

The certification empty weight and balance data shall include the unusable fuel and the total engine oil as specified:

- Airplane Flight Manual is required.

#### 2. Placards

All required placards as listed in the Pilatus Aircraft Flight Manual, and subsequent approved revisions, must be installed in the appropriate locations.

#### 3. Continued Airworthiness

- Airworthiness Limitations are contained in Chapter 4 of the Pilatus AMM. These Limitations may not be changed without EASA approval.
- Current weight and balance data together with a list of equipment included in the certificated empty weight, and loading instructions, when necessary, must be provided for each airplane at the time of original certification.
- The basic variant PC-12 (S/N 101 – 683) may be converted to a variant PC-12/45 by executing Pilatus Service Bulletin No. 04-001.
- Only interior configurations described in the official Pilatus AFM/POH are approved for installation in the PC-12, PC-12/45, PC-12/47 and PC-12/47E aircraft. These configurations have been shown to meet the dynamic and HIC test requirements of FAR 23.562. Any alterations to these approved interior layouts must be shown to meet FAR 23.562.

#### 4. OSD

In the absence of an operational evaluation at the entry into service of the PC-12, a type-rating requirement was established by the JAA. With the outcome of the OSD-FCD aircraft evaluation from 02 - 06 Nov 2015 summarized in Pilatus Report PC-12 No 02394 all the PC-12 are thereafter subject to a Class Rating (prior this date it was a Type Rating) and OSD-FCD is no longer applicable to the PC-12.

Issue 07, 21 December 2016

**Acronyms**

AMM - Aircraft Maintenance Manual  
 CRI - Certification Review Item  
 FAR - Federal Aviation Regulations  
 FOCA - Federal Office of Civil Aviation  
 EASA - European Aviation Safety Agency  
 IAS - Indicated Airspeed  
 KIAS - Indicated Airspeed [knots]  
 MAC - Mean Aerodynamic Chord  
 POH - Pilot's Operating Handbook  
 RPM - Rotations per Minute  
 FIKI - Flight Into Known Icing  
 SLD - Supercooled Large Droplets  
 TCDS - Type Certificate Data Sheet

**II. Type Certificate Holder Record**

Pilatus Aircraft Ltd.  
 P.O. Box 992, 6371 Stans  
 Switzerland

**III. Change Record**

Revision	Date of revision	Change Description
Issue 01	29 June 2006	Initial issue of EASA TCDS based on the Swiss FOCA TCDS F-56-30, Revision 10, dated December 14 <sup>th</sup> , 2005.
Issue 02	04 July 2006	PC-12/47 eligible S/N 684 and up. Typo corrected.
Issue 03	28 March 2008	PC-12/47E eligible S/N 545 and MSN 1001 and up.
Issue 04	04.Nov.2009	Added a Note 5 for the RVSM capability of PC-12/47E
Issue 05	11.Feb 2014	Corrected fine pitch tolerance and added alternative Nose and Main wheel tires. Corrected Typo for Vo at 4740kg.
Issue 06	12.Nov.2015	<p>Certification basis for PC-12/47E updated indicating FAR 23.1303 (c) at amdt 62 for the removal of non-stabilized compass (Whisky compass).</p> <p>Added the Hartzell 5-Blade composite propeller as a new installation on PC-12/47E.</p> <p>Added Note 6 for eLDG installation on PC-12/47E.</p> <p>Added Note 7 for the 5-Blade propeller, certification basis for removed standby compass and the PC-12/47E is T-PED tolerant.</p> <p>Added Note 8 for the Parker Nose- and Main wheels and brakes as an alternative on the PC-12/47E.</p> <p>Added section V to refer to Operational Suitability Data (OSD) including Note and effected CB.</p>

Issue 07, 21 December 2016

Issue 07	21 Dec 2016	The reference to the tyre manufactures and part numbers has been deleted since the approved manufacturers and tyre part numbers are controlled in the IPC. OSD FCD acc ref Note 4
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