

Service Bulletin No: 25-059

Ref No: 469

Modification No: EC-25-0067

ATA Chapter: 25

**EQUIPMENT AND FURNISHINGS - PASSENGER COMPARTMENT
IMPROVEMENT TO THE PSU TRIM PANEL INSTALLATION ABOVE THE EMERGENCY EXIT****1. Planning Information****A. Effectivity**

PC-12/47E series aircraft MSN 2001 thru MSN 2999.

NOTE: Some aircraft were fitted with the modification proposed in this SB during production by the aircraft OEM. This SB does not need to be implemented if visual inspection confirms that the aircraft is already equipped with the dual locks. Refer to Figure 1.

B. Concurrent requirements

None.

C. Reason**(1) Problem**

During an inspection it was identified that it is possible for the PSU trim panel (installed above the emergency exit door) to block the opening of the emergency exit. This is possible if the PSU trim panel is positioned too far inboard due to incorrect installation, which can cause interference with the emergency exit door panel.

(2) Solution

A modification is necessary to improve the installation of the interior trim panels in the area of the emergency exit.

D. Description

This Service Bulletin gives the data and instructions necessary to:

- Part 1 - Determine if the dual lock fasteners are installed on the PSU trim panel
- Part 2 - If necessary, introduce additional fasteners on the interior trim panels in the area of the emergency exit.

E. Compliance**Mandatory.**

To be accomplished at the next scheduled annual inspection after the issue of this Service Bulletin.

F. Approval

The technical content of this document is approved under the authority of the DOA ref. EASA.21J.357.

Pilatus advises Operators/Owners to check with their designated Airworthiness Authority for any changes, local regulations or sanctions that may affect the embodiment of this Service Bulletin.

G. Copyright and legal statement

© Pilatus Aircraft Ltd. This document contains proprietary information that is protected by copyright. All rights are reserved. No part of this document may be copied, reproduced or translated to other languages without the prior written consent of Pilatus Aircraft Ltd.

In connection with the use of this document, Pilatus does not provide any express or implied warranties and expressly disclaims any warranty of merchantability or fitness for a particular purpose. This document contains trade secrets, confidential and/or proprietary information of Pilatus and technical data subject to export control laws and regulations, including the U.S. Export Administration Regulations (EAR). Disclosure or distribution of this document contrary to the EAR, and other laws and regulations, is strictly forbidden. The above restrictions may apply to data on all pages of this document.

H. Manpower

Description	Man-Hours	
	Part 1	Part 1 and Part 2
Preparation		0.5
Part 1 - Examination for installation of dual lock fasteners	0.25	1.25
Part 2 - Modification to install the dual lock fasteners	-	1.0
Requirements after job completion		0.5
TOTAL MAN-HOURS	1.25	2.25

NOTE: The man-hours do not include the time necessary to cure sealants, paints and adhesives.

I. Weight and balance

Not changed.

J. Electrical load change data

Not changed.

K. Software

Not changed.

L. References

AMM references are given without the model prefix:
For PC-12/47E Series aircraft use the prefix 12-B, and/or 12-C as applicable for
MSN 2001 thru MSN 2999.

Aircraft Maintenance Manual (AMM)

20-31-00-00A-070A-A

25-21-03-00A-920A-A

52-20-00-00A-920A-A

M. Publications affected

Aircraft Maintenance Manual (AMM).

Illustrated Parts Data (IPD).

N. Interchangeability of parts

One-way interchangeable.

2. Material Information

A. Material - Price and availability

Operators that require additional information and/or Service Bulletin material can contact their authorized Pilatus Service Center, or Pilatus Customer Support on:

<https://www.pilatus-aircraft.com> → contact us.

NOTE: Part numbers given in this Service Bulletin are correct at the time of approval. Pilatus Aircraft Ltd reserves the right to change the part numbers as necessary. Part numbers of items delivered are correct when dispatched. This could lead to differences between those part numbers quoted in this Service Bulletin and the delivered parts, if parts are superseded. Operators are requested to check the Illustrated Parts Data (IPD) for delivered parts that differ from those listed in the Service Bulletin Materials List.

Operators are requested to advise Pilatus Aircraft Ltd of the Manufacturer’s Serial Number (MSN), the flying hours and landings of aircraft that are allocated for this Service Bulletin.

B. Warranty

Credit will be issued for labour for all affected aircraft on approval of a warranty claim, provided the work is accomplished by an authorized Service Center within the compliance time given in Section 1.E on Page 1 of this Service Bulletin.

C. Material necessary for each aircraft

NOTE: Referenced AMM procedures may contain additional materials to be procured.

(1) Material to order from Pilatus

None

(2) Material to procure locally

New Part No.	Description	Old Part No.	Qty	Disp. Code	Fig.	Item
917.47.28.046	Fastener self-adhesive, Dual-lock (SUPPLIER P/N SJ3552CF)	-	AR	N	1	4
917.47.28.045	Fastener self-adhesive, Dual-lock (SUPPLIER P/N SJ3551CF)	-	AR	N	1	6

Disposition Codes: D - Discard / N - New / R - Return to Pilatus / E - Exchange Part

D. Locally supplied consumables

NOTE: Referenced AMM procedures may contain additional consumables to be used.

To identify the materials used in this procedure, look in the Consumable materials list.
Refer to AMM 12-B-20-31-00-00A-070A-A.

Material No.	Description	Qty	Remarks
P01-033	Wipes, presaturated solvent	AR	-
P08-080	Adhesive	AR	Scotch-Weld 10

E. Material necessary for each spare

Not applicable.

F. Re-identified parts

Not applicable.

G. Tools and equipment

NOTE: Referenced AMM procedures may contain additional tools to be used.

Tools and equipment	Recommended Pilatus part
Tool kit, mechanic	Local supply

3. Accomplishment Instructions

WARNING: BE CAREFUL WHEN YOU USE THE CONSUMABLE MATERIALS. OBEY THE MANUFACTURER'S HEALTH AND SAFETY INSTRUCTIONS AND ALL THE APPLICABLE LOCAL INSTRUCTIONS. CONSUMABLE MATERIALS CAN BE DANGEROUS AND CAN CAUSE DEATH OR INJURY TO PERSONNEL AND / OR DAMAGE TO EQUIPMENT.

NOTE: Refer to the manufacturer's instructions for the necessary time for consumable materials to cure/dry.

A. Preparation

- (1) Move the aircraft into a hangar.
- (2) Open the passenger or cargo door to get access to the modification area.
- (3) Remove the emergency exit door. Refer to AMM 12-B-52-20-00-00A-920A-A.

B. Part 1 - Examination for installation of dual lock fasteners
Refer to Figure 1

- (1) Remove and keep the PSU, trim panel RH (1) from over the emergency exit door in the passenger compartment. Refer to AMM 12-C-25-21-03-00A-920A-A.
- (2) Do a visual inspection for the installation of the dual-lock fastener tapes (4) and (6) on the panels that follow:
Refer to views C, E, G, and J.
 - FWD, Upper, RH (3)
 - AFT, Upper, RH (2)
 - PSU, trim panel RH (1).
- (a) If the dual-lock fastener tapes (4) and (6) are installed, install the PSU, trim panel RH (1) that you kept.
Refer to AMM 12-C-25-21-03-00A-920A-A.
 - 1 Make sure that the PSU, trim panel RH (1) is installed correctly and the dual-lock fastener tape (4) and (6) are engaged.
 - 2 Continue with the 'Requirements after job completion' and 'Documentation' sections of this Service Bulletin.
- (b) If the dual-lock fastener tapes (4) and (6) are not installed go to Part 2 of this Service Bulletin, refer to Para 3.C.

C. Part 2 - Modification to install the dual lock fasteners

- (1) Cut two pieces of the dual-lock fastener tape (4) (P/N 917.47.28.046) to 20 x 25 mm (0.79 x 0.98 in).
- (2) Cut two pieces of the dual-lock fastener tape (6) (P/N 917.47.28.045) to 13 x 13 mm (0.51 x 0.51 in).
- (3) If necessary, remove the panel cover material (leather/vinyl) in the installation area of the dual-lock fastener tape (6). Refer to views G and J.

NOTE: The dual-lock fastener tape is to be bonded to the structure of the composite panels.

- (4) Use wipes, presaturated solvent (Material No. P01-033) to clean the installation positions of the dual-lock fastener tapes (4) and (6) on the panels that follow:
Refer to views C, E, G, and J.
 - FWD, Upper, RH (3)
 - AFT, Upper, RH (2)
 - PSU, trim panel RH (1).
- (5) Let the solvent dry.
- (6) Apply a thin layer of adhesive (Material No. P08-080) to the mating surfaces of the dual-lock fastener tapes (4) and (6) and the panels that follow:
Refer to views C, E, G, and J.
 - FWD, Upper, RH (3)
 - AFT, Upper, RH (2)
 - PSU, trim panel RH (1).
- (7) Let the adhesive cure as per the manufacturers instructions.
- (8) Install the dual-lock fastener tape (4) and (6) on the mating surfaces of the panels that follow:
Refer to views C, E, G, and J.
 - FWD, Upper, RH (3)
 - AFT, Upper, RH (2)
 - PSU, trim panel RH (1).
 - (a) Make sure that the dual-lock fastener tape (4) on the PSU, trim panel RH (1) is aligned with the locator pins (5). Refer to views E and C.
 - (b) Make sure that the dual-lock fastener tape (6) is aligned with the edge of the panels (2) and (3) and 30mm (1.18 in) above the emergency exit. Refer to views G and J.
 - (c) Make sure that the dual-lock fastener tape (4) and (6) is securely fastened to the panels.

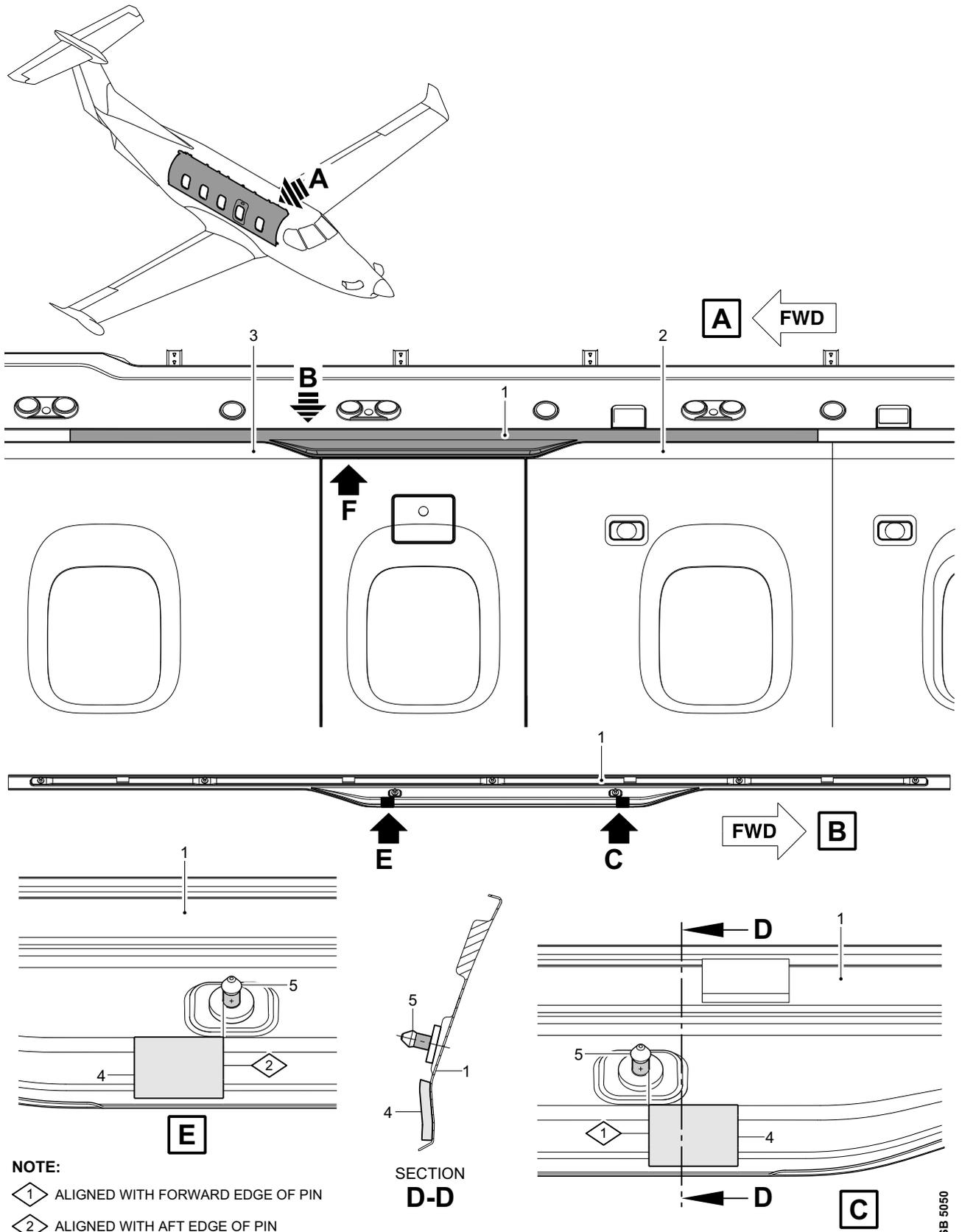
- (9) Install the PSU, trim panel RH (1) over the emergency exit door in the RH side of the passenger compartment. Refer to AMM 12-C-25-21-03-00A-920A-A.
- (10) Make sure that the PSU, trim panel RH (1) is installed correctly and the dual-lock fastener tape (4) and (6) are engaged.

D. Requirements after job completion

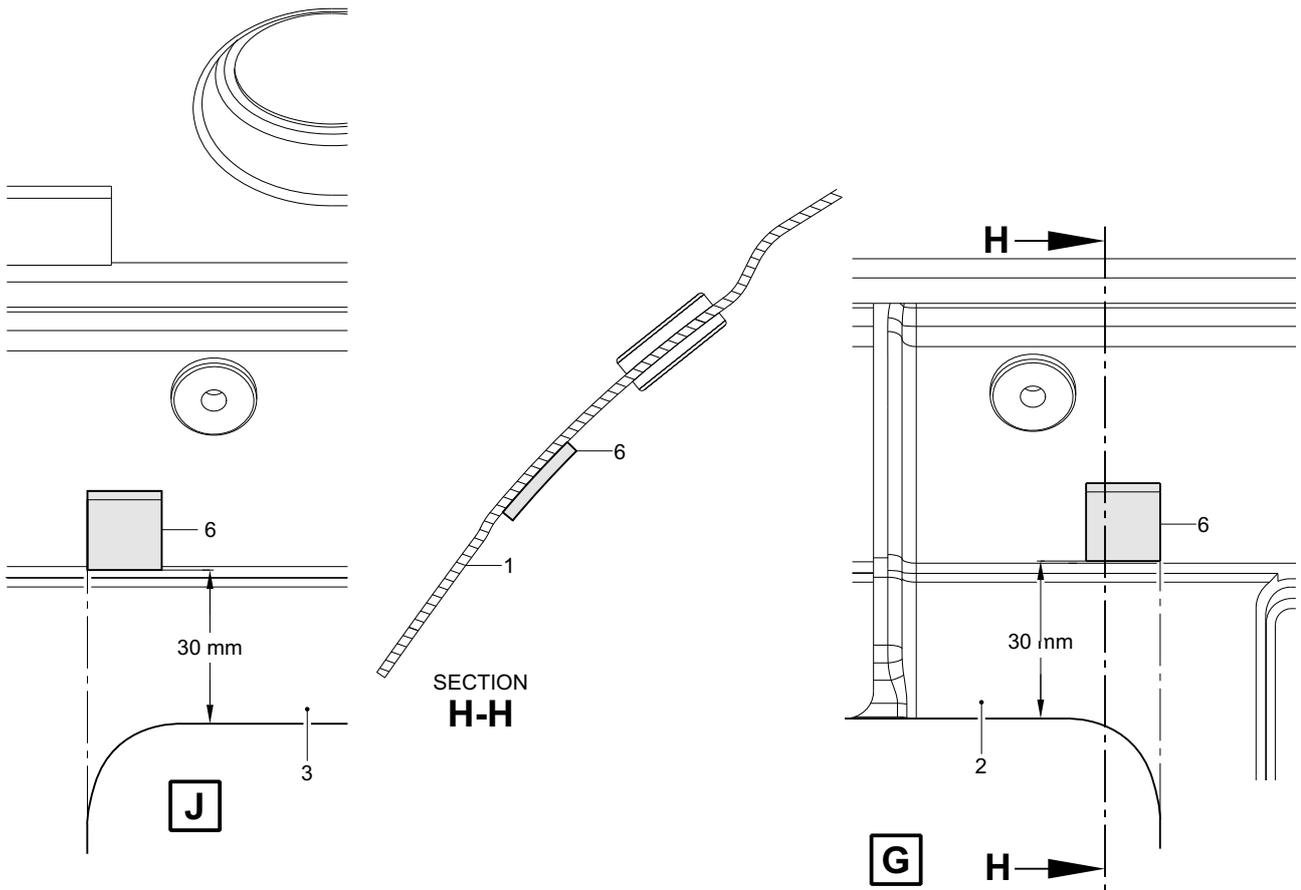
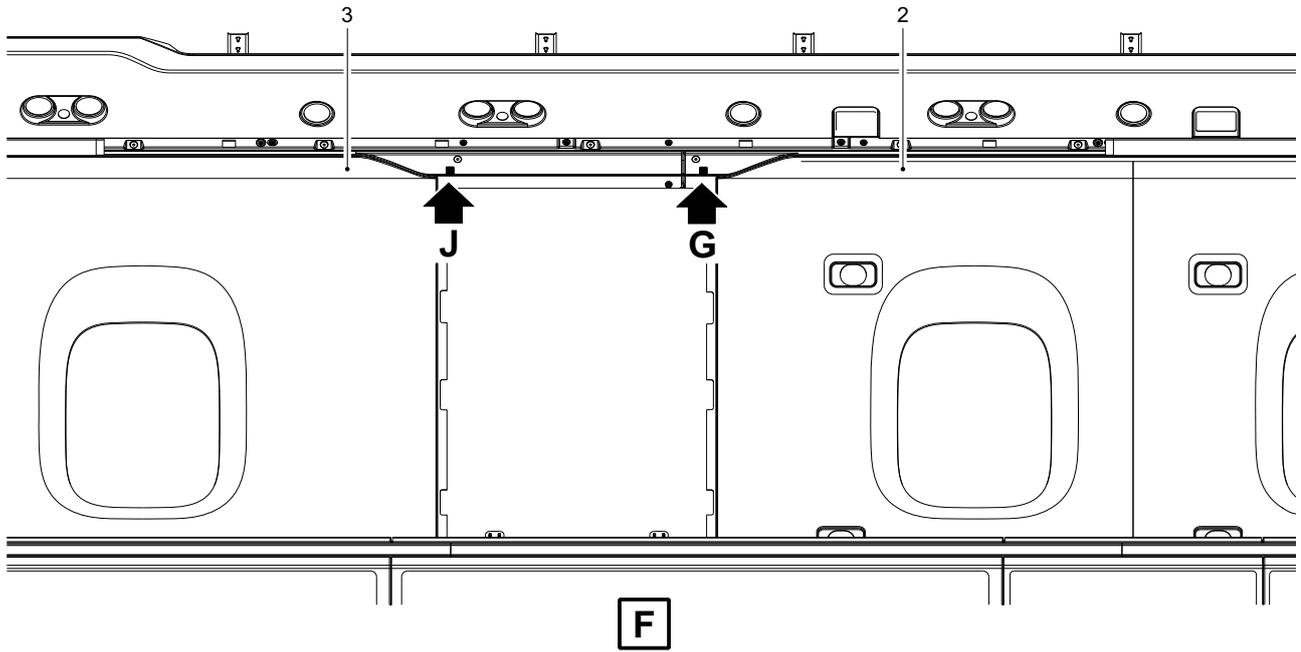
- (1) Install the emergency exit door. Refer to AMM 12-B-52-20-00-00A-920A-A.
- (2) Make sure that the PSU, trim panel RH (1) has sufficient clearance with the emergency exit door. Refer to Figure 1.
- (3) Remove all tools and materials. Make sure that the work areas are clean.

E. Documentation

- (1) Make an entry in the Aircraft Logbook to record the accomplishment of Part 1 only or Part 1 and Part 2 of this Service Bulletin.
- (2) Make sure that the Aircraft Logbook shows any new Pilatus Part Number(s) and/or Serial Number(s), as applicable.
- (3) Make an entry in the PC-12/47E Pilot's Operating Handbook, Document No. 02406, to record the incorporation of this Service Bulletin.
- (4) Make sure that all applicable Aircraft Documentation is updated.
- (5) Inform CAMP of the incorporation of this Service Bulletin and any new Pilatus Part Number(s) and/or Serial Number(s), as applicable. Send the completed CAMP feedback sheet to: fax@campsystems.com



Installation of dual-lock fastener tape
Figure 1 (Sheet 1 of 2)



SB 5051

Installation of dual-lock fastener tape
Figure 1 (Sheet 2 of 2)

Feedback Sheet for the Accomplishment of SB 25-059

The purpose of this feedback sheet is to provide CAMP with the current information on each individual PC-12 aircraft.

Please complete the grey cells as appropriate using black ink and block letters.

Print out and send the completed feedback sheet to: fax@campsystems.com

Aircraft MSN		Aircraft Registration		Total Airframe Hours	
Service Center				Total Landings	

SB Accomplishment Information

We have embodied/accomplished this SB		<input type="checkbox"/>	Fully	<input type="checkbox"/>	Partially
<input type="checkbox"/>	Part 1 only	<input type="checkbox"/>	Part 1 and Part 2		
The undersigned confirms the accomplishment of this Service Bulletin					
Date of accomplishment		Name		Signature	
Comments (procedure, kit quality, suggested improvements etc.)					

CAMP Feedback Sheet

INTENTIONALLY BLANK